Amendments to the Claims:

1. (currently amended) A method for repairing a turbine component, the method comprising the steps of:

cold gas-dynamic spraying powder material to buildup degraded areas on the turbine component; and

post-spray processing the turbine component <u>while heating the turbine component above a cold gas-dynamic spraying temperature</u> to consolidate applied material and restore metallurgical integrity to the repaired turbine component.

2. (original) The method of claim 1 wherein turbine component comprises a turbine blade.

3. (original) The method of claim 2 wherein the turbine blade comprises a tip, and wherein the cold gas-dynamic spraying is performed on the tip.

4. (original) The method of claim 2 wherein the turbine blade comprises a leading edge, and wherein the cold gas-dynamic spraying is performed on the leading edge.

5. (original) The method of claim 2 wherein the turbine blade comprises a platform, and wherein the cold gas-dynamic spraying is performed on the platform.

6. (original) The method of claim 2 wherein the turbine blade comprises a z-notch shroud, and wherein the cold gas-dynamic spraying is performed on the z-notch shroud.

7. (original) The method of claim 1 wherein the step of post-spray processing the turbine component to consolidate applied material and restore metallurgical integrity to the repaired turbine component comprises performing a vacuum sintering on the turbine component after the step of cold gas-dynamic spraying particles.

8. (original) The method of claim 7 wherein the step of performing a vacuum sintering comprises sintering in a vacuum for between 2 and 4 hours at temperatures of between 2050 degrees F and 2300 degrees F.

9. (original) The method of claim 1 wherein the step of post-spray processing the turbine component to consolidate applied material and restore metallurgical integrity to the repaired turbine component comprises performing a hot isostatic pressing on the turbine component after the step of cold gas-dynamic spraying particles.

10. (currently amended) The method of claim 9 wherein the step of performing a hot isostatic pressing on the turbine component comprises pressing for between 2 and 4 hours at temperatures of between 2100 and ~~23000~~<u>2300</u> degrees F and at pressures of between 10 and 30 ksi.

11. (original) The method of claim 9 further comprising the step of performing a rapid cooling of between 45 and 60 degrees F per minute to a desired temperature level after the hot isostatic pressing.

12. (original) The method of claim 1 wherein the step of post-spray processing the turbine component to consolidate applied material and restore metallurgical integrity to the repaired turbine component comprises a heat treatment of between 2 to 4 hours at temperatures of between 2000 and 2200 degrees F followed by a second heat treatment of between 16 to 24 hours at temperatures of between 1300 and 1800 degrees F.

13. (currently amended) A method for repairing a high pressure turbine component, the method comprising the steps of:

cold gas-dynamic spraying powder material to repair degradation in at least a portion of the high pressure turbine component;

vacuum sintering the high pressure turbine component <u>after the cold gas-dynamic spraying step</u>;

performing a hot isostatic pressing on the high pressure turbine component <u>after the vacuum sintering step</u>; and

heat treating the high pressure turbine component <u>after performing the hot isostatic pressing step</u>.

14. (original) The method of claim 13 wherein the high pressure turbine component comprises a turbine blade.

15. (original) The method of claim 14 wherein the turbine blade comprises a tip, and wherein the cold gas-dynamic spraying is performed on the tip.

16. (original) The method of claim 14 wherein the turbine blade comprises a leading edge, and wherein the cold gas-dynamic spraying is performed on the leading edge.

17. (original) The method of claim 14 wherein the turbine blade comprises a platform, and wherein the cold gas-dynamic spraying is performed on the platform.

18. (original) The method of claim 14 wherein the turbine blade comprises a z-notch shroud, and wherein the particles are sprayed on the z-notch shroud.

19. (original) The method of claim 13 wherein the step of performing a vacuum sintering comprises sintering in a vacuum for between 2 and 4 hours at temperatures of between 2050 degrees F and 2300 degrees F.

20. (currently amended) The method of claim 13 wherein the step of performing a hot isostatic pressing on the turbine component comprises pressing for between 2 and 4 hours at temperatures of between 2100 and ~~23000~~<u>2300</u> degrees F and at pressures of between 10 and 30 ksi.

21. (original) The method of claim 13 further comprising the step of performing a rapid cooling of between 45 and 60 degrees F per minute to a desired temperature level after the hot isostatic pressing.

22. (original) The method of claim 13 wherein the step heat treating the high pressure turbine component comprises a heat treatment of between 2 to 4 hours at temperatures of between 2000 and 2200 degrees F followed by a second heat treatment of between 16 to 24 hours at temperatures of between 1300 and 1800 degrees F.

23. (currently amended) A method for repairing degradation on a turbine blade, the method comprising the steps of:

providing repair powder material;

mixing the repair powder material into a flow of gas, the gas at a temperature below a melt temperature of the repair powder material;

accelerating the repair powder material mixed into the flow of gas; ~~and~~

directing the accelerated repair powder material to a target surface on the turbine blade, wherein the repair powder material deforms on the target surface to repair degradation on the turbine blade;

vacuum sintering the turbine blade for between 2 and 4 hours at temperatures of between 2050 degrees F and 2300 degrees F;

performing a hot isostatic pressing on the turbine blade for between 2 and 4 hours at temperatures of between 2100 and ~~23000~~2300 degrees F and at pressures of between 10 and 30 ksi; and

heat treating the turbine blade between 2 to 4 hours at temperatures of between 2000 and 2200 degrees F followed by a second heat treatment of between 16 to 24 hours at temperatures of between 1300 and 1800 degrees F.